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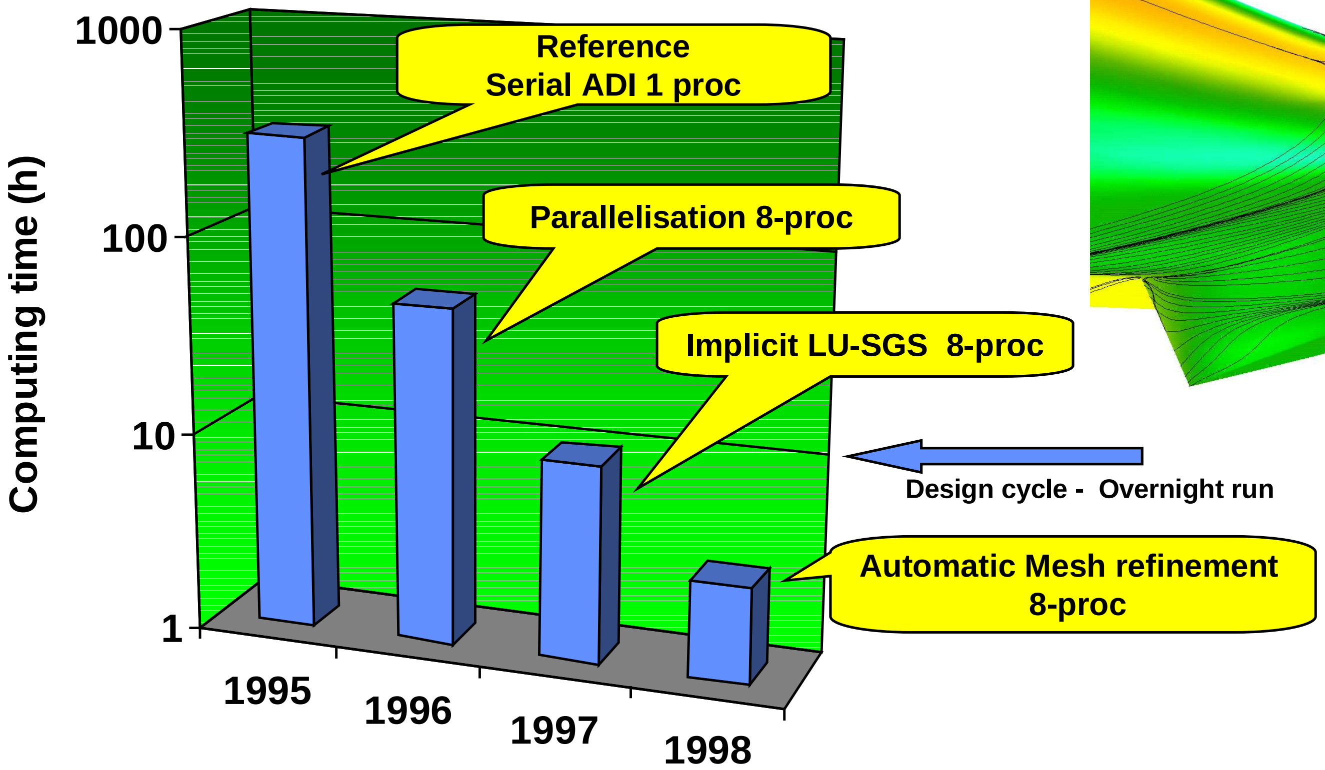
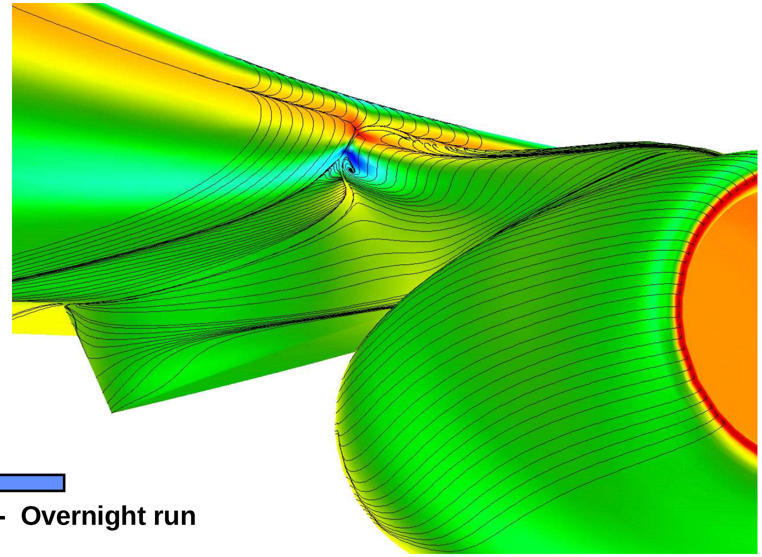
Navier-Stokes - multiblock algebraic turbulence model
Fuselage-Wing-Pylon-Nacelles - 65 blocks - 3,5 million pts
CRAY J916





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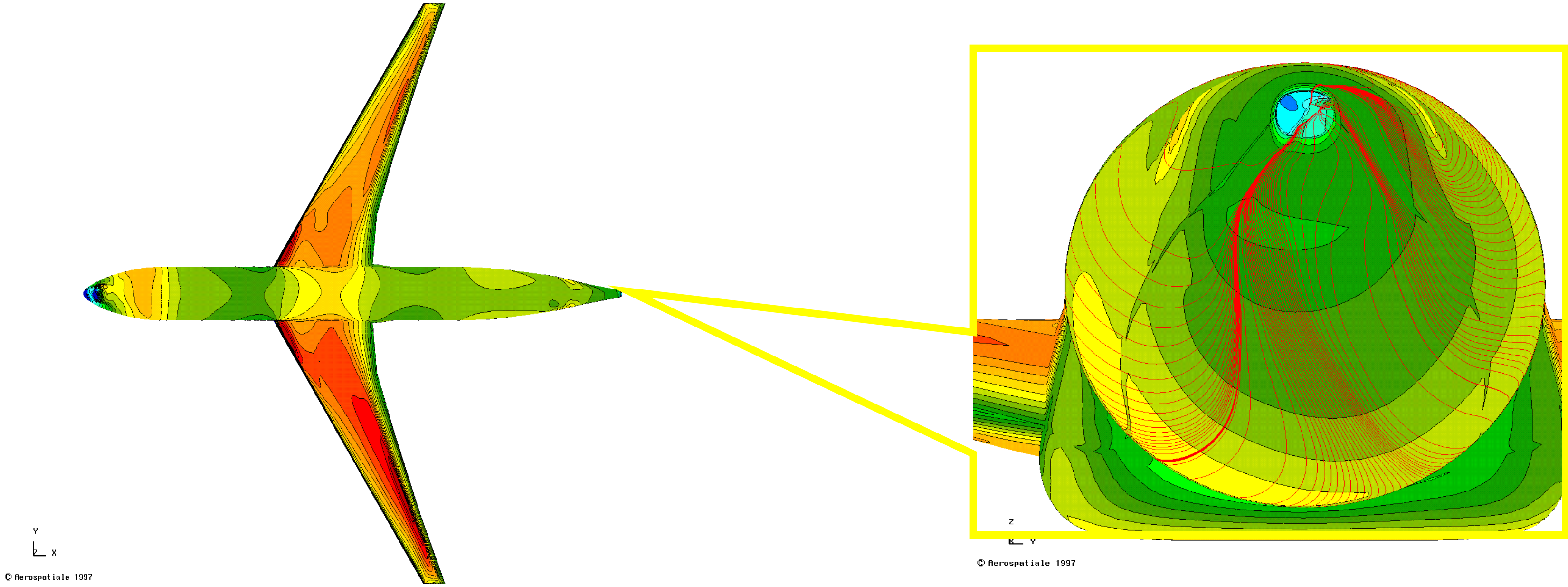
➢Integration of numerical and experimental means

⇨ **define experimental set-up and supplement experimental measurements**

Ex : rear fuselage flow analysis without any sting effects at sideslip conditions (Mach = 0.8 β= 4° , 3 millions grid points, 2000 iter., 8h elapsed time)



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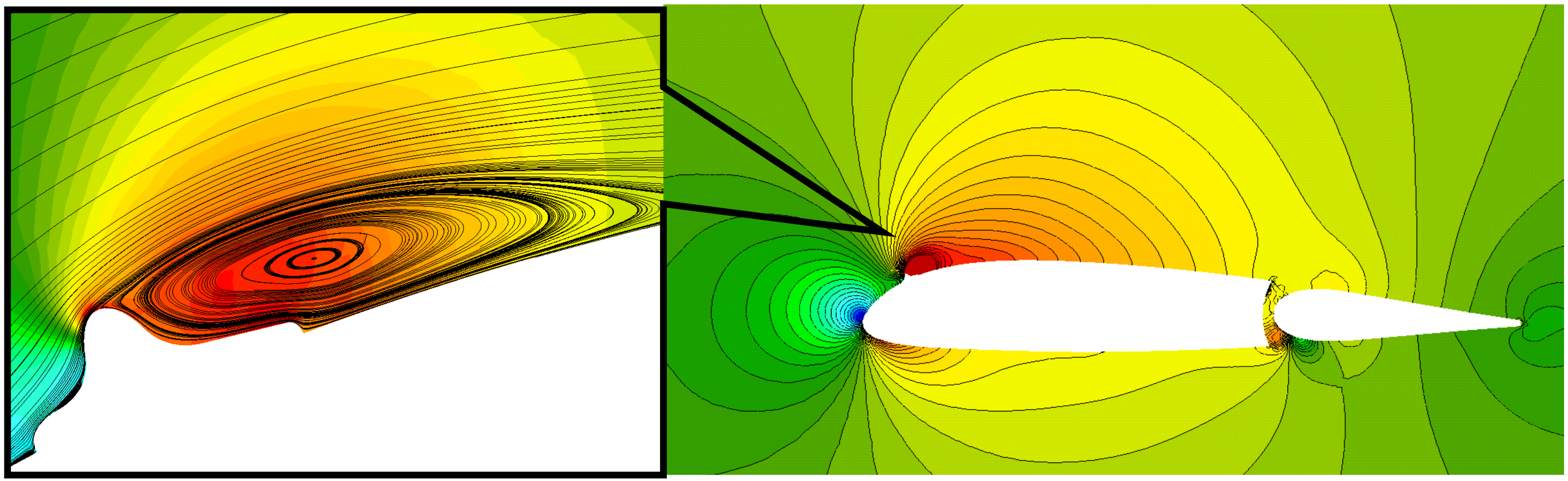
➢Aerodynamics interactions

⇨ **Icing effect on Aerospatiale-A airfoil with aileron at Mach 0.2 α= 0° , 155000 grid points, 2000 iter., 1.2h elapsed time**



Streamlines behind the Icing shape